WHAT IS CLAIMED IS:

1. A turbine blade to be installed into an engaged member of a turbine disk of an aircraft engine comprising:

a blade one side of which having a convex suction surface and the other side of which having a concave pressure surface;

a platform integrally molded on a hub side of the blade, a recess being formed on one side of the platform, a front seal fin formed protruding forward at the front end of the platform, and a rear seal fin formed protruding backward at the back end of the platform;

an engagement member integrally molded on the hub side of the platform, the engagement member having a engagement face which is able to be engaged with the engaged member and is formed by grinding;

a front engagement member integrally molded in the vicinity of a base portion of the front seal fin, the front engagement member having a front engagement face able to engage with a front locating portion of a jig to be used for the grinding, and the front engagement face located back from a virtual plane including the one side of the platform;

a front wall integrally molded in the vicinity of the base portion of the front seal fin, the front wall surrounding a front side-edge portion of the front engagement member;

a rear engagement member integrally molded in the vicinity of a base portion of the rear seal fin, the rear engagement member having a rear engagement face able to engage with a rear locating portion of the jig, and the rear engagement face located back from the virtual plane; and

a rear wall integrally molded in the vicinity of the base portion of the rear seal fin, the rear wall surrounding a rear side-edge portion of the rear engagement member,

wherein an end face of the front wall and an end face of the

rear wall are respectively configured to be coplanar with the virtual plane.

2. The turbine blade of claim 1, wherein the front engagement face and the rear engagement face are respectively configured to be substantially parallel to the longitudinal direction of the engagement member.

3. The turbine blade of claim 1, wherein the spacing between the front edge of the front engagement face and the rear edge of the rear engagement face is configured to be longer than the longitudinal length of the engagement member.

4. The turbine blade of claim 2, wherein the spacing between the front edge of the front engagement face and the rear edge of the rear engagement face is configured to be longer than the longitudinal length of the engagement member.

5. The turbine blade of claim 1, wherein each of the front engagement face and the rear engagement face has a recess in a range of less than or equal to 0.7 mm.

6. The turbine blade of claim 2, wherein each of the front engagement face and the rear engagement face has a recess in a range of less than or equal to 0.7 mm.

7. The turbine blade of claim 3, wherein each of the front engagement face and the rear engagement face has a recess in a range of less than or equal to 0.7 mm.

8. The turbine blade of claim 4, wherein each of the front engagement face and the rear engagement face has a recess in a range

of less than or equal to 0.7 mm.

9. The turbine blade of claim 1, wherein the engaged member is a female dovetail and the engagement member is a male dovetail.